



No.	CF-2011-44R1	1/2
Issue Date	1 February 2012	

AIRWORTHINESS DIRECTIVE

The following airworthiness directive (AD) may be applicable to an aircraft which our records indicate is registered in your name. ADs are issued pursuant to **Canadian Aviation Regulation (CAR) 521 Division X**. Pursuant to **CAR 605.84** and the further details of **CAR Standard 625, Appendix H**, the continuing airworthiness of a Canadian registered aircraft is contingent upon compliance with all applicable ADs. Failure to comply with the requirements of an AD may invalidate the flight authorization of the aircraft. Alternative means of compliance shall be applied for in accordance with **CAR 605.84** and the above-referenced **Standard**.

This AD has been issued by the Continuing Airworthiness Division (AARDG), National Aircraft Certification Branch, Transport Canada, Ottawa, telephone 613 952-4357.

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**TRANSPORT CANADA EMERGENCY AIRWORTHINESS DIRECTIVE
PLEASE FORWARD IMMEDIATELY TO THE PERSON RESPONSIBLE FOR THE
OPERATION AND MAINTENANCE OF YOUR AIRCRAFT**

Number: CF-2011-44R1

Subject: Main Rotor Blade

Effective: Upon Receipt

Revision: Supersedes Airworthiness Directive CF-2011-44

Applicability: Bell Helicopter Textron Canada Models 206L, L-1, L-3 and L-4 helicopters equipped with a main rotor blade part number 206-015-001-107, -109, -111, -115, -117, -119 or -121.

Compliance: As indicated below, unless already accomplished.

Background: An investigation into a Model 206L-1 accident has revealed that a main rotor blade has fractured as a result of fatigue. This Airworthiness Directive (AD) was originally issued with the understanding that the fatigue crack may occur if the following conditions are present: residual stress in the blade's spar and a significant void in the adhesive between the inertia weight and the spar, between stations 100 and 145. The presence of a crack in the spar could lead to the total failure of the blade and the loss of the helicopter.

Subsequent investigation into a second accident on a Model 206L helicopter has led to the following conclusions:

- Presently, there is no reliable inspection method to detect the crack on the blade before blade failure, and
- To insure continued airworthiness, a reduced life limit is necessary, irrespective of the size of an adhesive void.

Revision R1 of this AD introduces a life limit of 1400 hours air time on all affected main rotor blades regardless of previous inspections performed.

Corrective Actions: **Part I – Verification**

Before next flight from the effective date of this AD, verify the part number and the serial number of each main rotor blade to determine if the blade is affected by this AD, in accordance with the Accomplishment Instructions Part I of the Bell Alert Service Bulletin (ASB) 206L-09-159 Rev. A dated 13 November 2009, or later revisions approved by the Chief, Continuing Airworthiness, Transport Canada.

Pursuant to **CAR 202.51** the registered owner of a Canadian aircraft shall, within seven days, notify the Minister in writing of any change of his or her name or address.

To request a change of address, contact the **Civil Aviation Communications Centre (AARC)** at Place de Ville, Ottawa, Ontario K1A 0N8, or 1-800-305-2059, or www.tc.gc.ca/civilaviation/communications/centre/address.asp

